

AD-A138 143

CREEP OF CARBON YARNS AND COMPOSITES AT HIGH
TEMPERATURE(U) AEROSPACE CORP EL SEGUNDO CA MATERIALS
SCIENCES LAB L A FELDMAN 01 DEC 83 TR-0084(4728-02)-1
SD-TR-83-78 F04701-83-C-0084

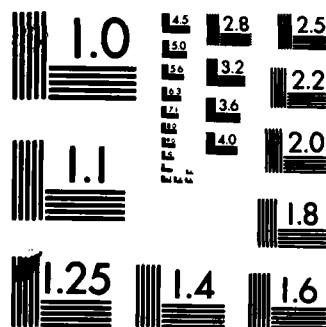
1/1

UNCLASSIFIED

F/G 11/5

NL

END



MICROCOPY RESOLUTION TEST CHART
NATIONAL BUREAU OF STANDARDS-1963-A

12

AD A138143

Creep of Carbon Yarns and Composites at High Temperature

L. A. FELDMAN
Materials Sciences Laboratory
Laboratory Operations
The Aerospace Corporation
El Segundo, California 90245

1 December 1983

DTIC
ELECTE
FEB 22 1984
B

APPROVED FOR PUBLIC RELEASE;
DISTRIBUTION UNLIMITED

DTIC FILE COPY

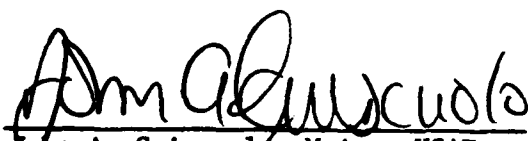
Prepared for
OFFICE OF NAVAL RESEARCH
Washington, D.C. 22217
SPACE DIVISION
AIR FORCE SYSTEMS COMMAND
Los Angeles Air Force Station
P.O. Box 92960, Worldway Postal Center
Los Angeles, California 90009

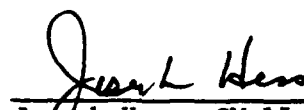
84 02 21 083

This report was submitted by The Aerospace Corporation, El Segundo, CA 90245, under Contract No. F04701-83-C-0084 with the Space Division, P.O. Box 92960, Worldway Postal Center, Los Angeles, CA 90009. It was reviewed and approved for The Aerospace Corporation by L. R. McCreight, Director, Materials Sciences Laboratory. Major John A. Criscuolo, West Coast Office, AF Space Technology Center, was the Air Force project officer.

This report has been reviewed by the Public Affairs Office (PAS) and is releasable to the National Technical Information Service (NTIS). At NTIS, it will be available to the general public, including foreign nationals.

This technical report has been reviewed and is approved for publication. Publication of this report does not constitute Air Force approval of the report's findings or conclusions. It is published only for the exchange and stimulation of ideas.


John A. Criscuolo, Major, USAF
Project Officer


Joseph Hess, GM-15, Director
West Coast Office, AFSTC

UNCLASSIFIED

SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)

REPORT DOCUMENTATION PAGE		READ INSTRUCTIONS BEFORE COMPLETING FORM
1. REPORT NUMBER SD-TR-83-78	2. GOVT ACCESSION NO. AD-A138143	3. RECIPIENT'S CATALOG NUMBER
4. TITLE (and Subtitle) CREEP OF CARBON YARNS AND COMPOSITES AT HIGH TEMPERATURE		5. TYPE OF REPORT & PERIOD COVERED
		6. PERFORMING ORG. REPORT NUMBER TR-0084(4728-02)-1
7. AUTHOR(s) L. A. Feldman		8. CONTRACT OR GRANT NUMBER(s) F04701-83-C-0084
9. PERFORMING ORGANIZATION NAME AND ADDRESS The Aerospace Corporation El Segundo, California 90245		10. PROGRAM ELEMENT, PROJECT, TASK AREA & WORK UNIT NUMBERS
11. CONTROLLING OFFICE NAME AND ADDRESS Office of Naval Research Washington, D.C. 22217		12. REPORT DATE 1 December 1983
14. MONITORING AGENCY NAME & ADDRESS (if different from Controlling Office) Space Division Los Angeles Air Force Station Los Angeles, California 90009		13. NUMBER OF PAGES 12
		15. SECURITY CLASS. (of this report) Unclassified
16. DISTRIBUTION STATEMENT (of this Report) Approved for public release; distribution unlimited.		15a. DECLASSIFICATION/DOWNGRADING SCHEDULE
17. DISTRIBUTION STATEMENT (of the abstract entered in Block 20, if different from Report)		
18. SUPPLEMENTARY NOTES		
19. KEY WORDS (Continue on reverse side if necessary and identify by block number) High-temperature creep Carbon matrix Carbon fiber X-ray diffraction Carbon yarn Carbon-carbon composite		
20. ABSTRACT (Continue on reverse side if necessary and identify by block number) Creep of mesophase- and rayon-based carbon yarns was measured in 2 to 4 ksi tension from 2100°C to failure (2500-2700°C). Resulting creep rates increased with temperature, and activation energies were lower for mesophase- than for rayon-based yarn. Rates of creep measured in pitch-matrix-impregnated yarn bundles were lower than those for unimpregnated bundles.		

DD FORM 1473
(IFACSIMILE)

UNCLASSIFIED

SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)

PREFACE

This work was supported by the Office of Naval Research under the direction of Dr. L. H. Peebles (Contract No. F04701-83-C-0084). The author thanks Dr. R. A. Meyer for initiating this effort and Dr. P. J. Blatz and Prof. G. A. Sines for helpful discussions. The valuable experimental assistance of D. C. Robinson is gratefully acknowledged.

S **DTIC**
ELECTE
FEB 22 1984 **D**
B

Accession For	
NTIS GRA&I	<input checked="checked" type="checkbox"/>
DTIC TAB	<input type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By _____	
Distribution/	
Availability Codes	
Avail and/or	
Dist	Special
A-1	



CONTENTS

PREFACE.....	1
I. INTRODUCTION.....	7
II. EXPERIMENTAL.....	9
III. RESULTS AND DISCUSSION.....	11
IV. CONCLUSIONS.....	15
REFERENCES.....	17

FIGURES

1.	SEM view of mesophase-pitch-impregnated P55 yarn after 2750°C under load.....	10
2.	Schematic of carbon-yarn creep experiment.....	10
3.	Typical creep behavior (observed experimentally) of unimpregnated yarn, increasing with temperature.....	12
4.	High-temperature failure samples.....	12
5.	Comparison of elongation rates for three yarn samples, showing marked difference between heat-treated and untreated yarns.....	13
6.	Thermally activated behavior of dry-yarn creep (elongation) rates.....	13
7.	Temperature dependence of impregnated P55 creep rate, showing marked departure from linearity.....	14

I. INTRODUCTION

Processing of carbon-carbon composites is affected both by the stresses that occur during pyrolysis and by the thermal expansion anisotropy of the reinforcing carbon fibers and matrix materials. Defects, such as delaminations in two-dimensional composites and cracks and debonds in three-dimensional composites, appear in the final structure. As the material is heated above 2000°C during processing, stresses begin to relax by means of creep (time-dependent deformation). The effects of creep on the stresses developed later on cooling have not been fully investigated, in part because of a lack of information about the creep behavior of the load-bearing fibers during composite processing. Some information has been presented in the literature on creep behavior of carbon fibers.¹⁻⁴ However, the modeling of composite behavior during processing has been based mainly on thermal expansion and elastic and fracture properties.⁵ The high-temperature creep behavior of some carbon fibers used in carbon-carbon composites is investigated here.

II. EXPERIMENTAL

Creep experiments were conducted using two types of Union Carbide carbon fibers in the form of multifilament yarns: VSB-32 (or P55), a mesophase-pitch-based carbon fiber consisting of 2000 filaments (10 μm average diameter) per yarn with nominal Young's modulus of 55 Msi; and WCA carbon cloth, from which continuous sections of yarn were removed. Also tested were simple carbon-carbon composite samples produced from yarns impregnated at 400°C with mesophase-transformed A240 pitch and calcined at 800°C under argon. A scanning electron microscope (SEM) photograph of a typical section of impregnated P55 yarn is presented in Fig. 1. Creep was measured at increasing temperatures above 2000°C, using a horizontal graphite tube furnace. As indicated in Fig. 2, the yarn sample was passed through the tube, after one end was anchored externally. The other sample end was attached to a flexible wire passing over a pulley and connected to a hanging weight at the other end. The experiment was conducted in an inert argon atmosphere inside the furnace. Displacement of the fiber end was detected by a linear variable differential transformer (LVDT). A feedback controller and automatic optical pyrometer regulated the temperature, and a digital data logger recorded data. Scanning electron microscopy, X-ray diffraction, and electrical resistance measurements also were used to determine the changes in fibers caused by high-temperature creep.

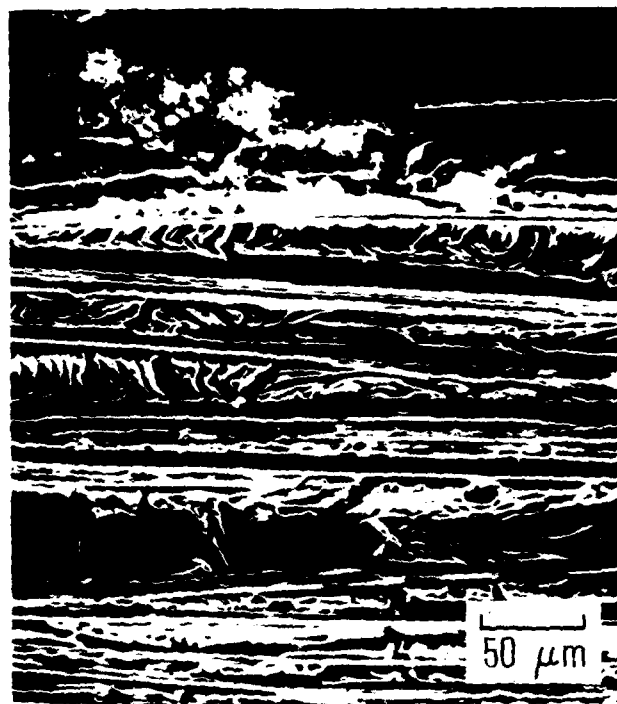


Fig. 1. SEM view of mesophase-pitch-impregnated P55 yarn after 2750°C under load.

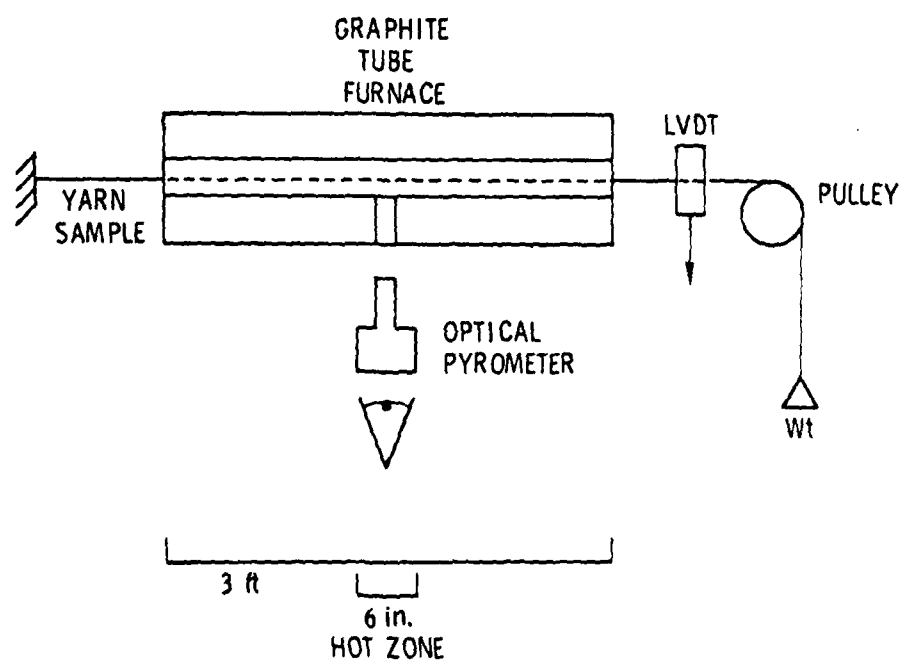


Fig. 2. Schematic of carbon-yarn creep experiment.

III. RESULTS AND DISCUSSION

A typical experimental result for creep, Fig. 3, indicates that creep rate increases steadily with temperature from 2060 to 2600°C, then necks down at 2600°C, as the inset illustrates for a WCA filament. Diameter reduction is greater than a factor of 3. Filaments became extremely narrow (0.1-0.5 μm) near the point of failure, separating into subfilaments over a region 50 to 100 μm in length (original fiber diameter $\sim 10 \mu\text{m}$) before the actual point of failure. A comparison of the failed ends of dry P55 filaments with pitch-matrix-impregnated P55 filaments, Fig. 4, reveals that the impregnated filaments have narrowed slightly (to $\sim 6\text{-}7 \mu\text{m}$) but have fractured ends with a rough, serrated appearance.

The observed creep strain rates were influenced by the heat-treatment history. Figure 5 compares typical elongation rates above 2000°C for P55 yarns, both as-received and heat-treated, and WCA yarn. The indicated elongation rates are significantly higher for the P55 as-received than the same material after heat treatment at 2600°C, which is consistent with the increase in graphitization of the material at high temperature. The elongation rate of the WCA yarn falls between the two P55 results. This result agrees with the fact that WCA cloth is carbonized from rayon cloth at high temperatures, so that further graphitization and hot-stretching in the experiment are not expected below the previous graphitization temperature.

An Arrhenius plot of log of elongation rate versus inverse temperature is roughly linear for the WCA and P55 dry yarns. The steady-state creep rate is typically given by a relation of the form²

$$\dot{\epsilon} = A \exp(-\Delta H/RT) \quad (1)$$

Average activation energies were 109 and 66 kcal/mol (see Fig. 6), which contrast with the behavior of the impregnated P55 material plotted in Fig. 7. A drop is observed in the elongation rate of the impregnated yarn sample at around 2400°C and may be caused by the continuing graphitization of the matrix

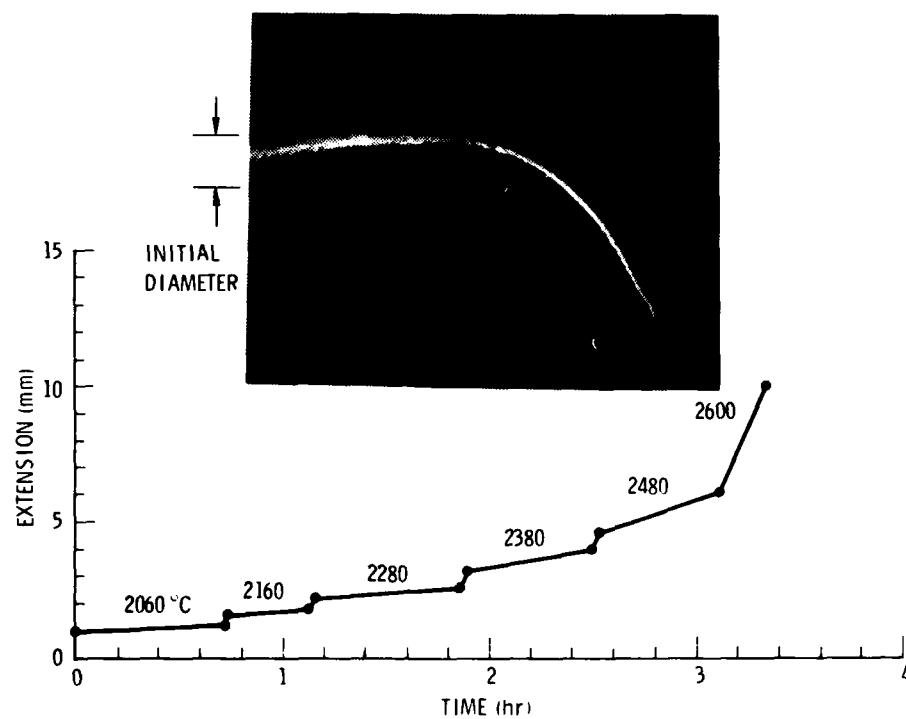


Fig. 3. Typical creep behavior (observed experimentally) of unimpregnated yarn, increasing with temperature. Inset shows failed end of typical WCA filament in hot zone.

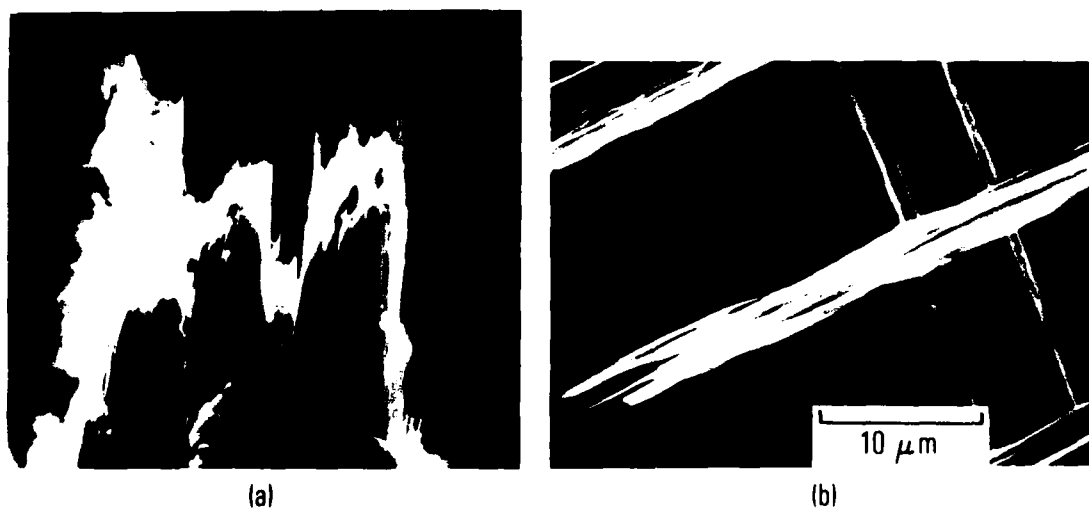


Fig. 4. High-temperature failure samples: (a) mesophase-pitch impregnated; (b) P55, dry.

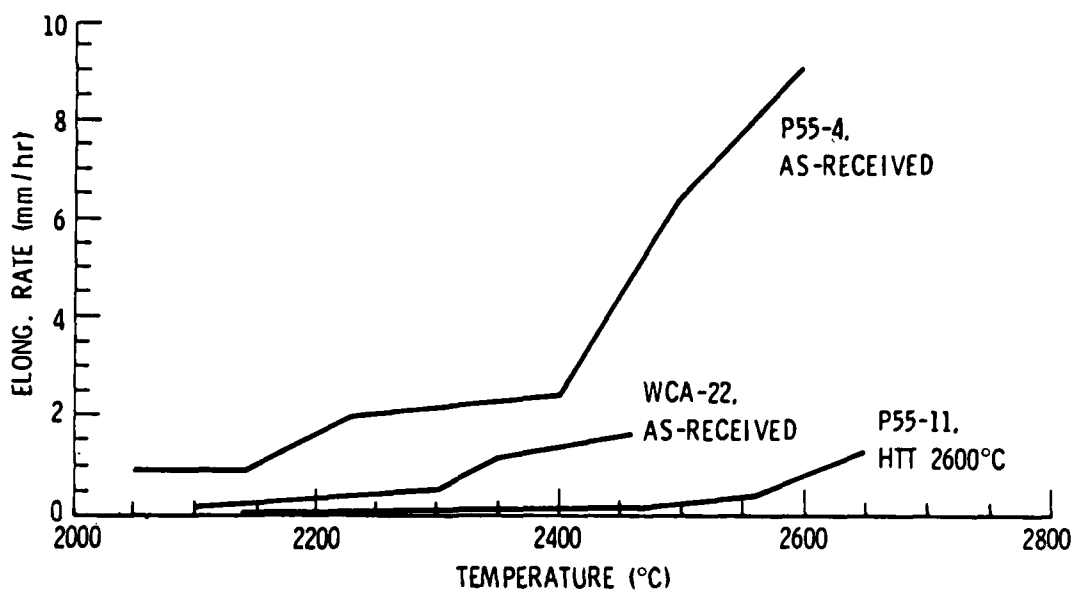


Fig. 5. Comparison of elongation rates for three yarn samples, showing marked difference between heat-treated and untreated P55 yarns.

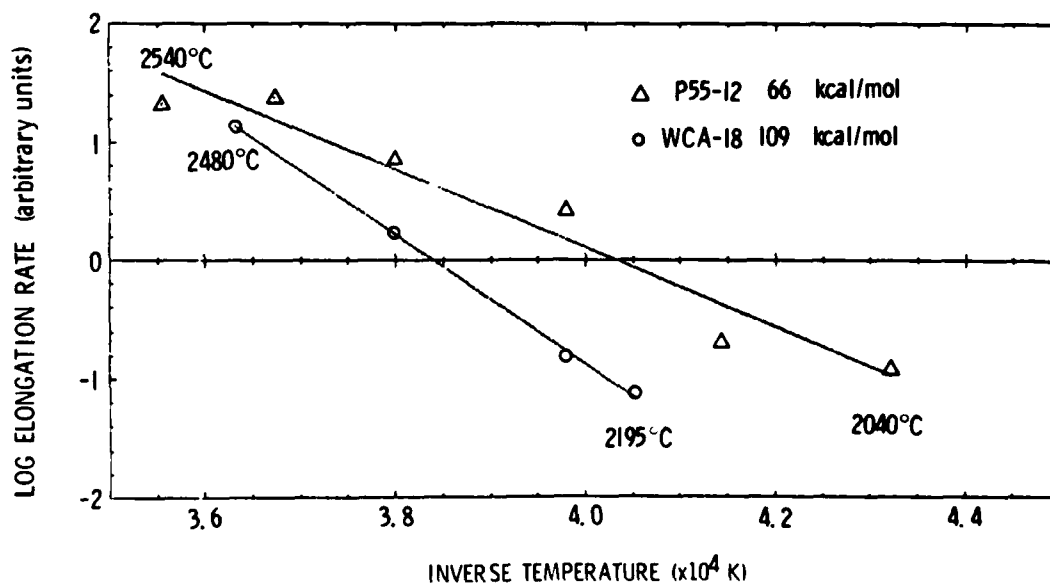


Fig. 6. Thermally activated behavior of dry-yarn creep (elongation) rates.

material and the contrasting thermomechanical properties of the matrix and filaments, though further work is needed to confirm the cause.

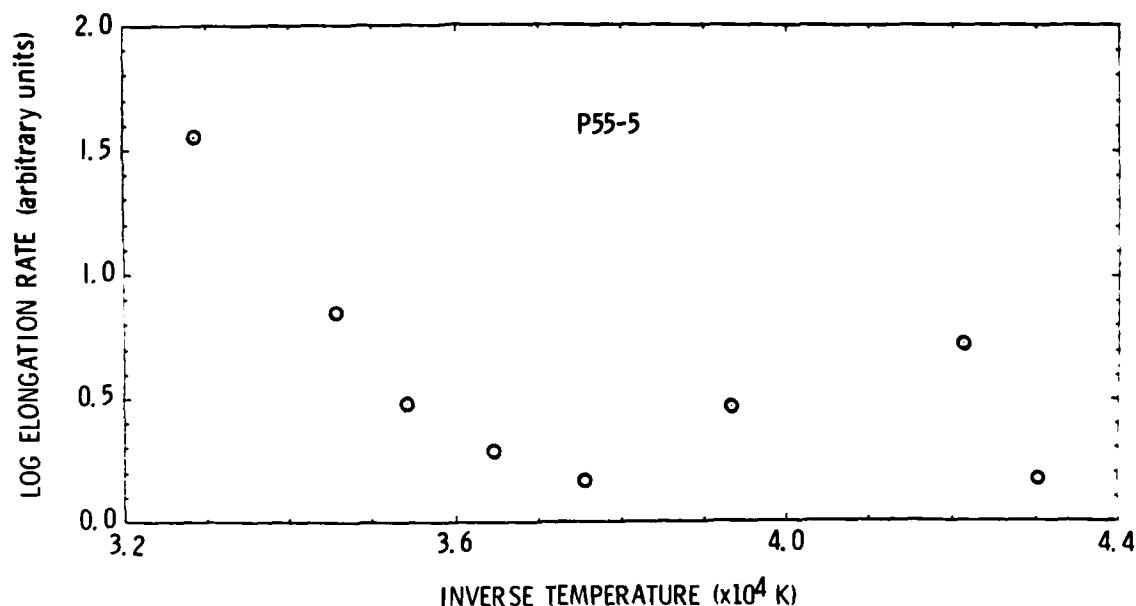


Fig. 7. Temperature dependence of impregnated P55 creep rate, showing marked departure from linearity.

X-ray diffraction revealed slight evidence that graphitization increases during heating, but not as a result of the applied stress. Unstressed fiber samples heated along with creep specimens exhibited similar changes in c-spacing and electrical resistivity. P55 yarn showed a decrease in the d(002) spacing from 0.3447 to 0.3392 nm; the WCA showed a decrease from 0.346 to 0.339 nm. Both c-spacings remained above the ideal graphite value of 0.335 nm.⁶ Measurements of the resistivity ratio $[\rho(77 \text{ K})/\rho(300 \text{ K})]$ increased from 1.12 to 1.67 for P55 after heat treatment at 2500°C, for both unstressed and creep samples. The WCA showed no change in resistivity ratio (less than 1%). X-ray diffraction indicated higher preferred orientation in P55 than WCA but little change in preferred orientation after heating or creep.

IV. CONCLUSIONS

Unimpregnated yarns of rayon-based and mesophase-pitch-based fibers demonstrated basic thermally activated creep behavior, whereas the creep-rate temperature dependence of matrix-impregnated yarns was more complex. Comparison of the failure behavior of samples indicated that necking was substantially less in impregnated than in unimpregnated yarns. Finally, the preferred orientation and crystallinity were altered by treatment at high temperature but were not affected by the degree of creep strain.

REFERENCES

1. R. Bacon and W. A. Smith, in 2nd Conf. on Industrial Carbon and Graphite, London (1965), pp. 203-213.
2. H. M. Hawthorne, J. Mat. Sci. 11, 97-110 (1976).
3. D. L. Lowe, Carbon Graphite Yarn Characterization - Carbon/Carbon Constituent Sensitivity Study, GE Re-Entry and Environmental Systems Division (1976).
4. C. D. Pears, "A Technical Note on Yarn Properties that Relate to Processing," Current Awareness Bulletin, Spec. Ed., Carbon-Carbon Composite Materials, No. 7, Metals and Ceramics Info. Ctr., Battelle Columbus Laboratories (25 June 1980).
5. W. K. Loomis, F. K. Tso, R. B. Dirling, Jr., and C. N. Heightland, Analytical Processing for Improved Composites (APIC), AFWAL-TR-81-4082 (1981).
6. W. N. Reynolds, Physical Properties of Graphite, Elsevier, New York (1968).

LABORATORY OPERATIONS

The Laboratory Operations of The Aerospace Corporation is conducting experimental and theoretical investigations necessary for the evaluation and application of scientific advances to new military space systems. Versatility and flexibility have been developed to a high degree by the laboratory personnel in dealing with the many problems encountered in the nation's rapidly developing space systems. Expertise in the latest scientific developments is vital to the accomplishment of tasks related to these problems. The laboratories that contribute to this research are:

Aerophysics Laboratory: Launch vehicle and reentry aerodynamics and heat transfer, propulsion chemistry and fluid mechanics, structural mechanics, flight dynamics; high-temperature thermomechanics, gas kinetics and radiation; research in environmental chemistry and contamination; cw and pulsed chemical laser development including chemical kinetics, spectroscopy, optical resonators and beam pointing, atmospheric propagation, laser effects and countermeasures.

Chemistry and Physics Laboratory: Atmospheric chemical reactions, atmospheric optics, light scattering, state-specific chemical reactions and radiation transport in rocket plumes, applied laser spectroscopy, laser chemistry, battery electrochemistry, space vacuum and radiation effects on materials, lubrication and surface phenomena, thermionic emission, photosensitive materials and detectors, atomic frequency standards, and bioenvironmental research and monitoring.

Electronics Research Laboratory: Microelectronics, GaAs low-noise and power devices, semiconductor lasers, electromagnetic and optical propagation phenomena, quantum electronics, laser communications, lidar, and electro-optics; communication sciences, applied electronics, semiconductor crystal and device physics, radiometric imaging; millimeter-wave and microwave technology.

Information Sciences Research Office: Program verification, program translation, performance-sensitive system design, distributed architectures for spaceborne computers, fault-tolerant computer systems, artificial intelligence, and microelectronics applications.

Materials Sciences Laboratory: Development of new materials: metal matrix composites, polymers, and new forms of carbon; component failure analysis and reliability; fracture mechanics and stress corrosion; evaluation of materials in space environment; materials performance in space transportation systems; analysis of systems vulnerability and survivability in enemy-induced environments.

Space Sciences Laboratory: Atmospheric and ionospheric physics, radiation from the atmosphere, density and composition of the upper atmosphere, aurorae and airglow; magnetospheric physics, cosmic rays, generation and propagation of plasma waves in the magnetosphere; solar physics, infrared astronomy; the effects of nuclear explosions, magnetic storms, and solar activity on the earth's atmosphere, ionosphere, and magnetosphere; the effects of optical, electromagnetic, and particulate radiations in space on space systems.

END

FILMED

384

DTIC